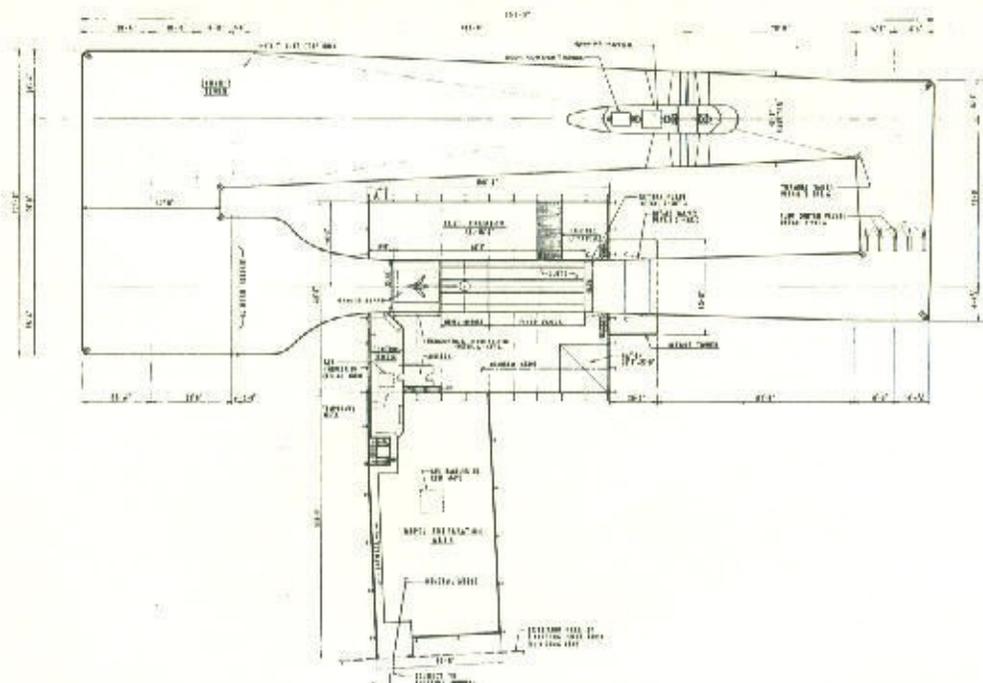


V/STOL TRANSITION RESEARCH WIND TUNNEL



This project will provide a research facility to obtain quantitative research information on the factors affecting the stability, control, and performance of V/STOL and STOL (short take-off and landing) aircraft. This facility is designed for frequent and easy model alterations, and will be used for general research investigations and provide adequate speed for investigations of the complete transition speed range at full-scale velocities and disc loadings.

Description: Single return closed wind tunnel powered by 8000 h.p. electric motor.

Test section 15 foot by 21 1/2 foot by 80 foot.

Moving ground board, with boundary layer control slot, has capability of speed up to 200 knots.

Data recording system.

Model support system is capable of:

- (a) Changing angle of attack through an angle of $\pm 45^\circ$.
- (b) Changing yaw angle through an angle of $\pm 45^\circ$.
- (c) Rotating the entire support on a turntable through an angle of 360° .

Status: Initial use is expected about July, 1968

Estimated Cost:

<u>Fiscal Year</u>	<u>C of F</u>	<u>Project Number</u>
1965	\$ 50,000	2356
1966	498,000	2356
1967	5,047,000	2356
Total =	\$5,595,000	